

3rd Annual Microgravity Environment Interpretation Tutorial

QUASI-STEADY ACCELERATION ONBOARD INTERNATIONAL SPACE STATION

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GRAVITY GRADIENT (1)



Considering a platform on a circular orbit. In the center of mass we have:

$$g_0 \frac{R_0^2}{R^2} = w^2 R$$

Where g_0 is the value of acceleration at sea level, R_0 is the Earth radius, w is the orbital angular velocity and R is the orbit radius.

The gravity gradient can be computed by considering a point located at a distance ℓ separately for the three cases along x, along y and along z

axes

1) ℓ along z

It is the simplest case and we have to substitute $(R \pm \ell)$ to R and carry out a linearization process

$$\begin{aligned} mg &= g_0 \frac{R_0^2}{(R \pm \ell)^2} - w^2 (R \pm \ell) = g_0 \frac{R_0^2}{R^2} \left(1 \pm \frac{\ell}{R} \right)^{-2} - g_0 \frac{R_0^2}{R^3} (R \pm \ell) = \\ &= g_0 \frac{R_0^2}{R^2} \left(1 \pm \frac{\ell}{R} \right)^{-2} - \left(1 \pm \frac{\ell}{R} \right) \cdot g_0 \frac{R_0^2}{R^2} \end{aligned}$$

GRAVITY GRADIENT (2)



2) ℓ along x

In this case you have to substitute $(R^2 + \ell^2)^{1/2}$ to R since R , ℓ and r form a rectangular triangle, where r is the hypotenuse:

$$mg = g_0 \frac{R_0^2}{R^2 + \ell^2} - \omega^2 (R^2 + \ell^2)^{\frac{1}{2}} = g_0 \frac{R_0^2}{R^2} \frac{\hat{E}}{\hat{E}} + \frac{\ell^2}{R^2} \frac{\hat{E}}{\hat{E}} - \frac{1}{2} \frac{\ell^2}{R^2} \frac{\hat{E}}{\hat{E}} + \frac{3}{2} g_0 \frac{R_0^2}{R^2} \frac{\ell^2}{R^2}$$

2) ℓ along y

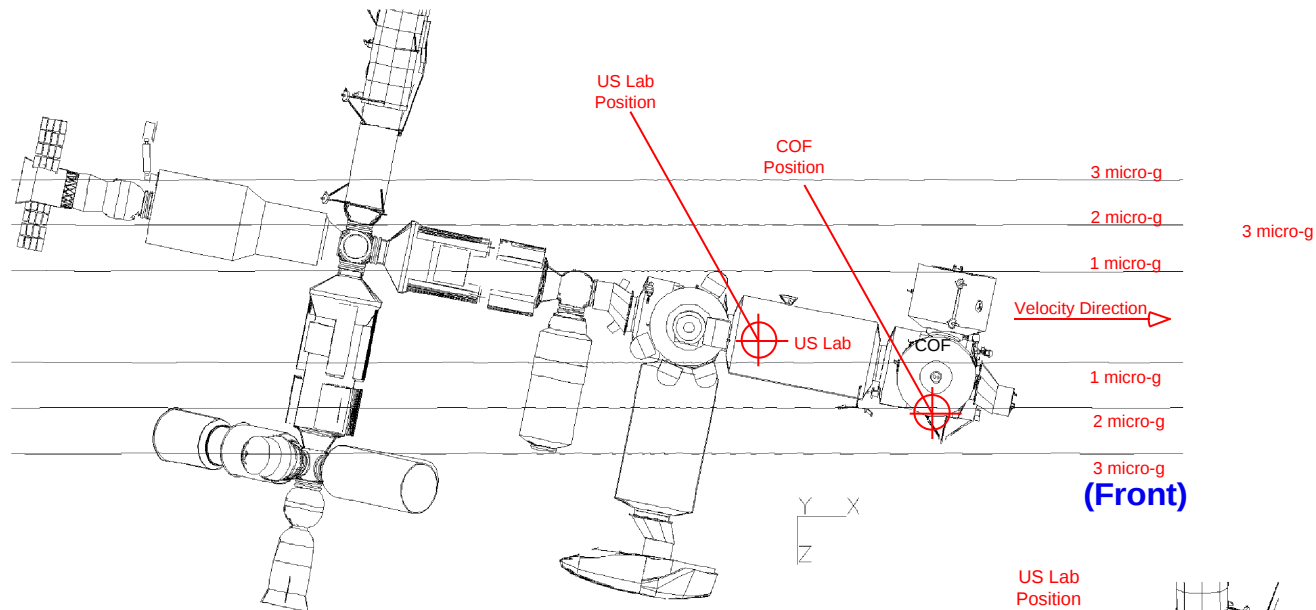
In this case the centrifugal force is the same. You have to project g along y taking into account that R , ℓ and r form a rectangular triangle, where r is the hypotenuse:

$$mg = g_0 \frac{R_0^2}{(R^2 + \ell^2)^{\frac{3}{2}}} (R\mathbf{k} + \ell\mathbf{j}) \cdot \mathbf{j} = g_0 \frac{R_0^2}{(R^2 + \ell^2)^{\frac{3}{2}}} \ell @ g_0 \frac{R_0^2}{R^2} \frac{\hat{E}}{\hat{E}} - \frac{3}{2} \frac{\ell^2}{R^2} \frac{\ell}{R} @$$

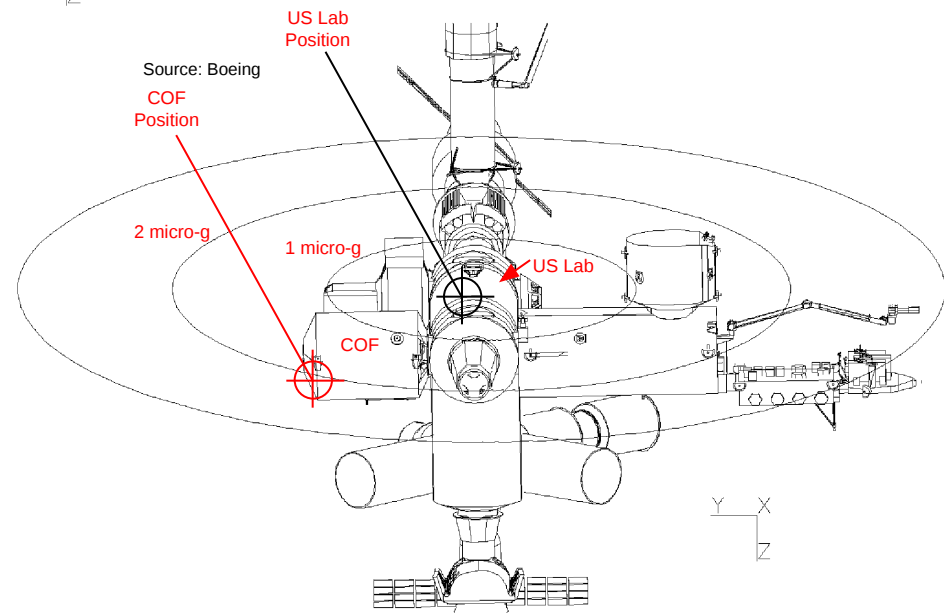
$$@ g_0 \frac{R_0^2}{R^2} \frac{\ell}{R}$$

DAC5 ASSEMBLY COMPLETE MICROGRAVITY ISO-G CONTOURS

(Side)

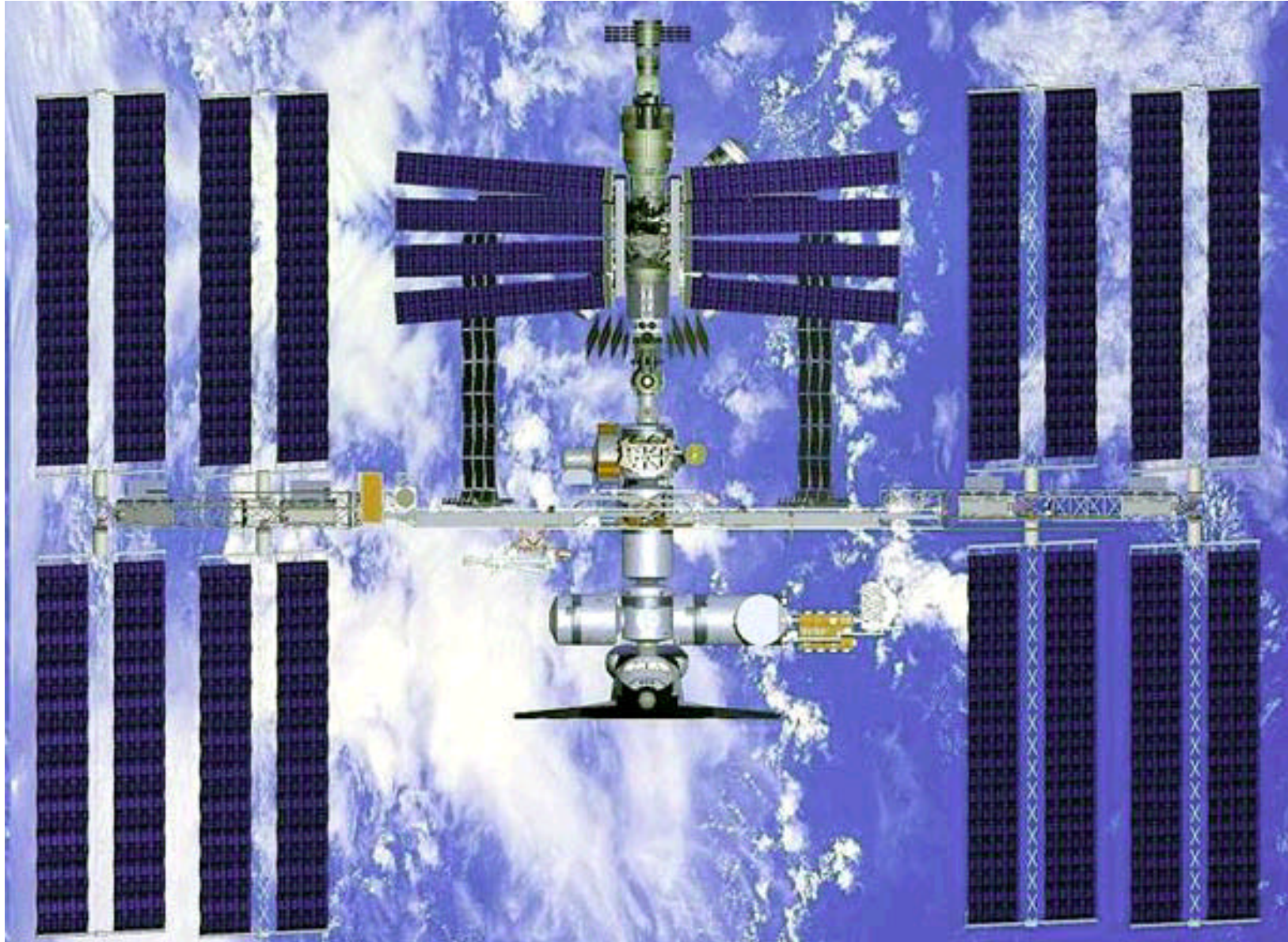


(Front)



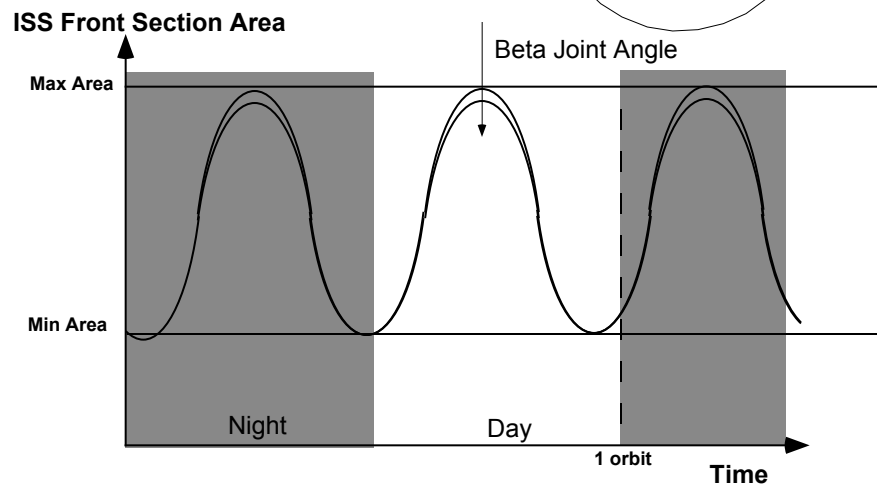
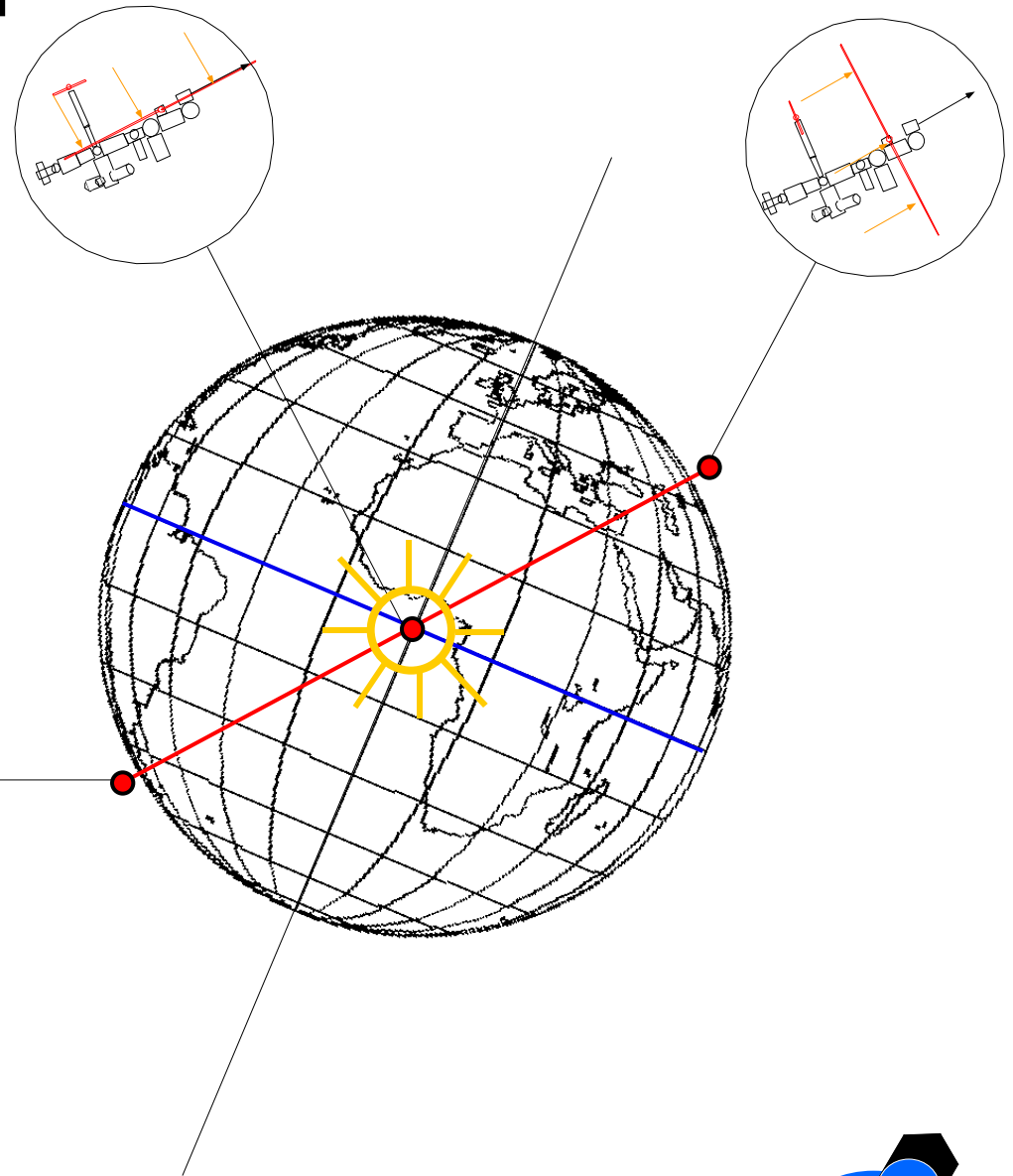
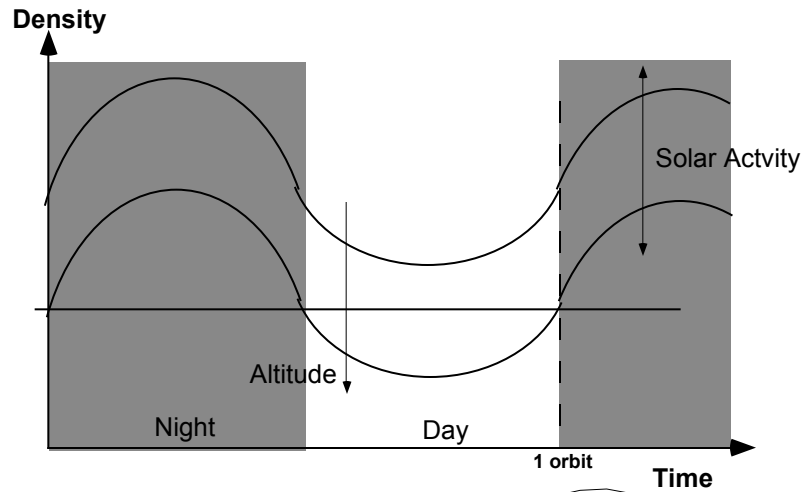
Source: Boeing

ISS Artistic View (Top)

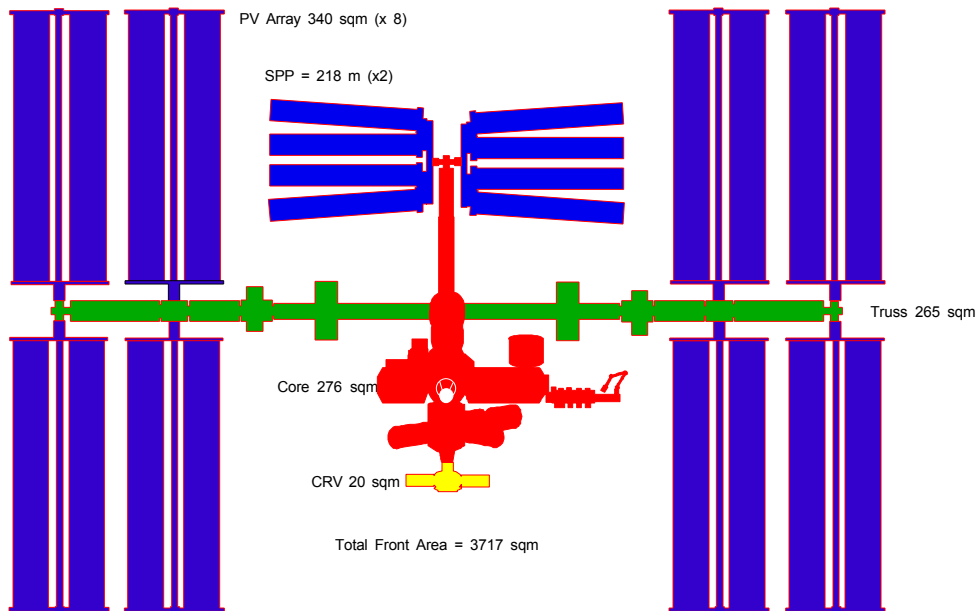


Source: Boeing

ISS SOLAR PANEL POSITION DURING ORBIT



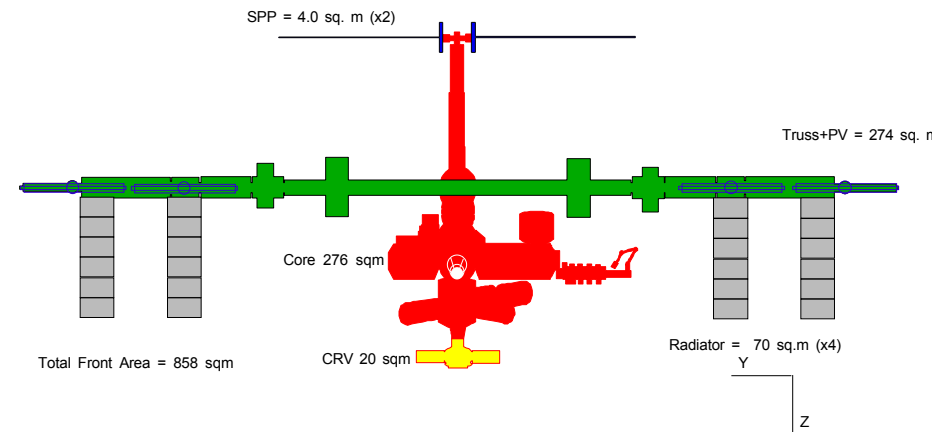
ISS RESIDUAL-g EVALUATION: aerodynamic drag contribution



maximum Drag:

Total Front Area = 3717 sqm
Altitude = 352.000 m
Velocity = 7694 m/s
Air Density = 7×10^{-12} Kg/m³
Cd = 2.5

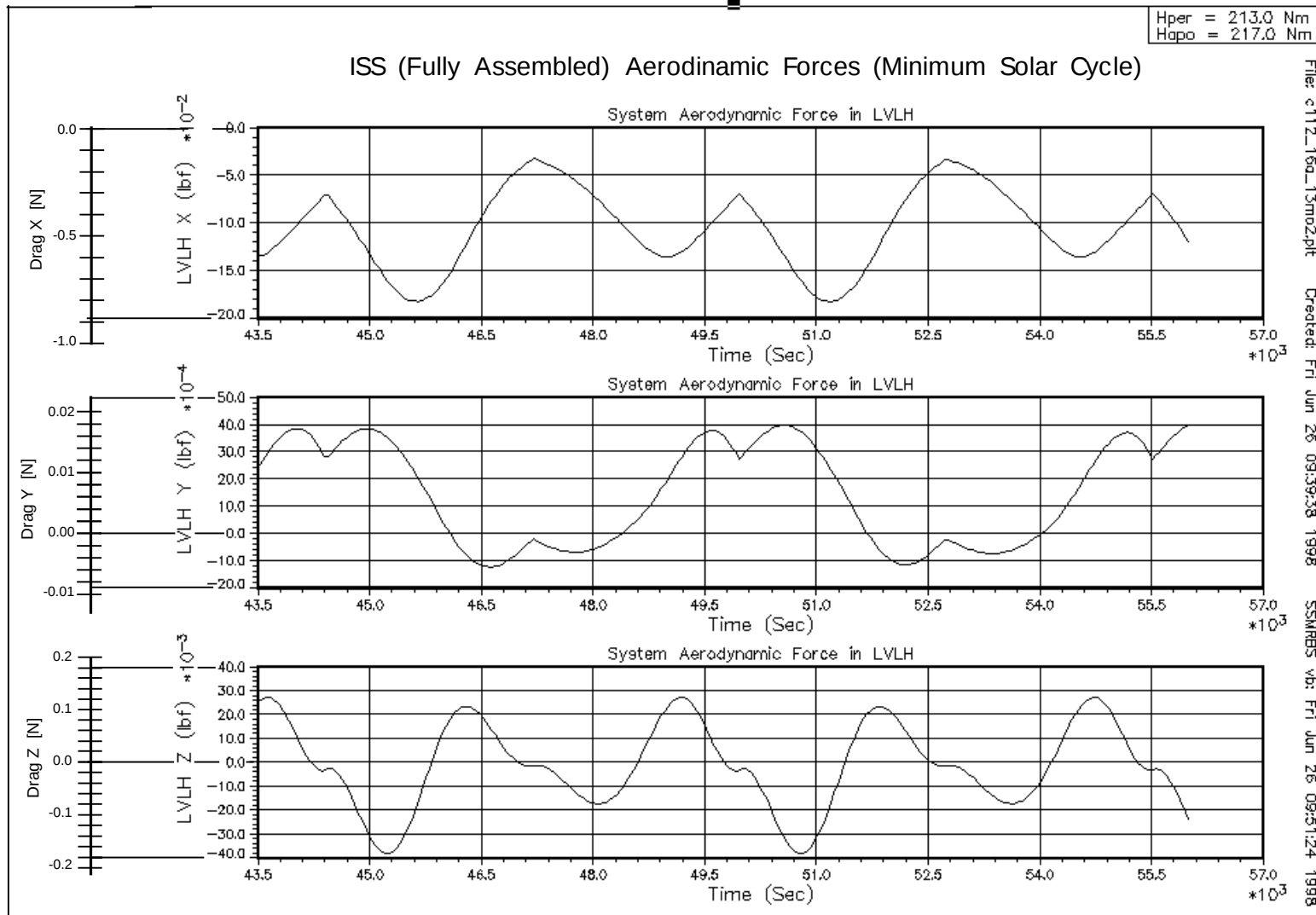
Drag X = 1.92 N
ISS Mass = 419.000 Kg
Acceleration X = 4.68×10^{-7} g₀



minumum Drag:

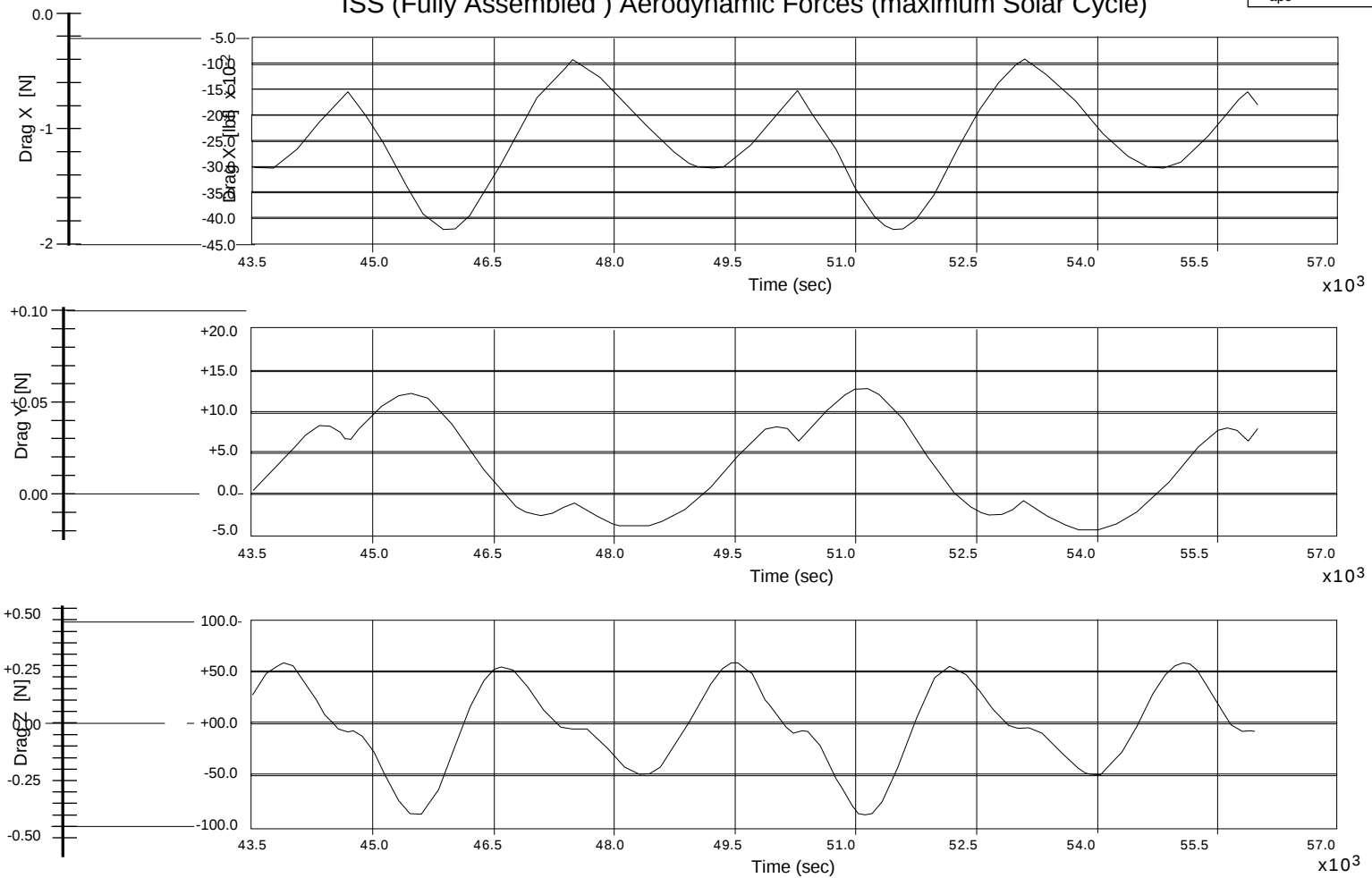
Total Front Area = 858 sqm
Altitude = 352.000 m
Velocity = 7694 m/s
Air Density = 7×10^{-12} Kg/m³
Cd = 2.5

Drag X = .44 N
ISS Mass = 419.000 Kg
Acceleration X = 1.1×10^{-7} g₀



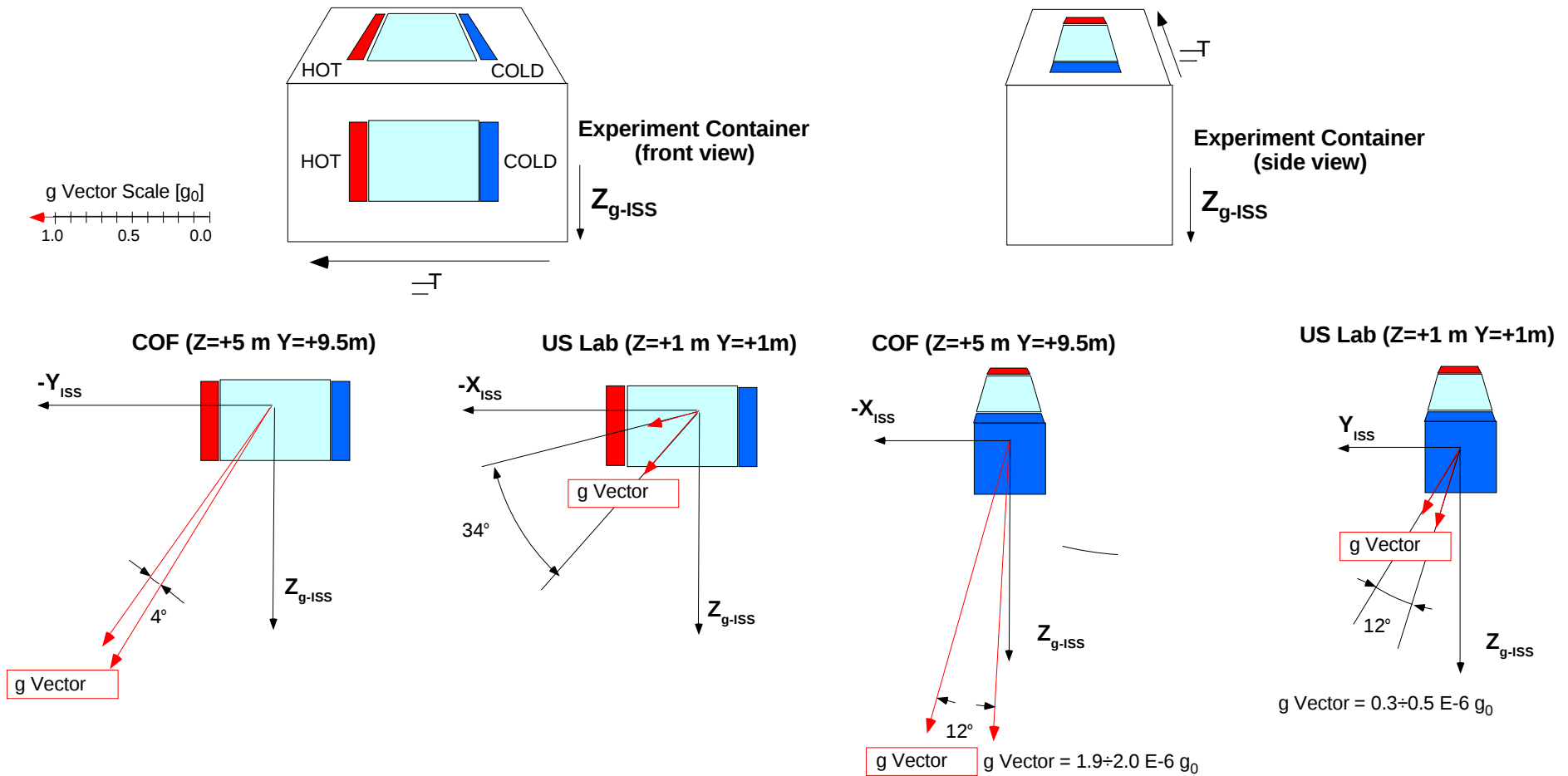
ISS (Fully Assembled) Aerodynamic Forces (maximum Solar Cycle)

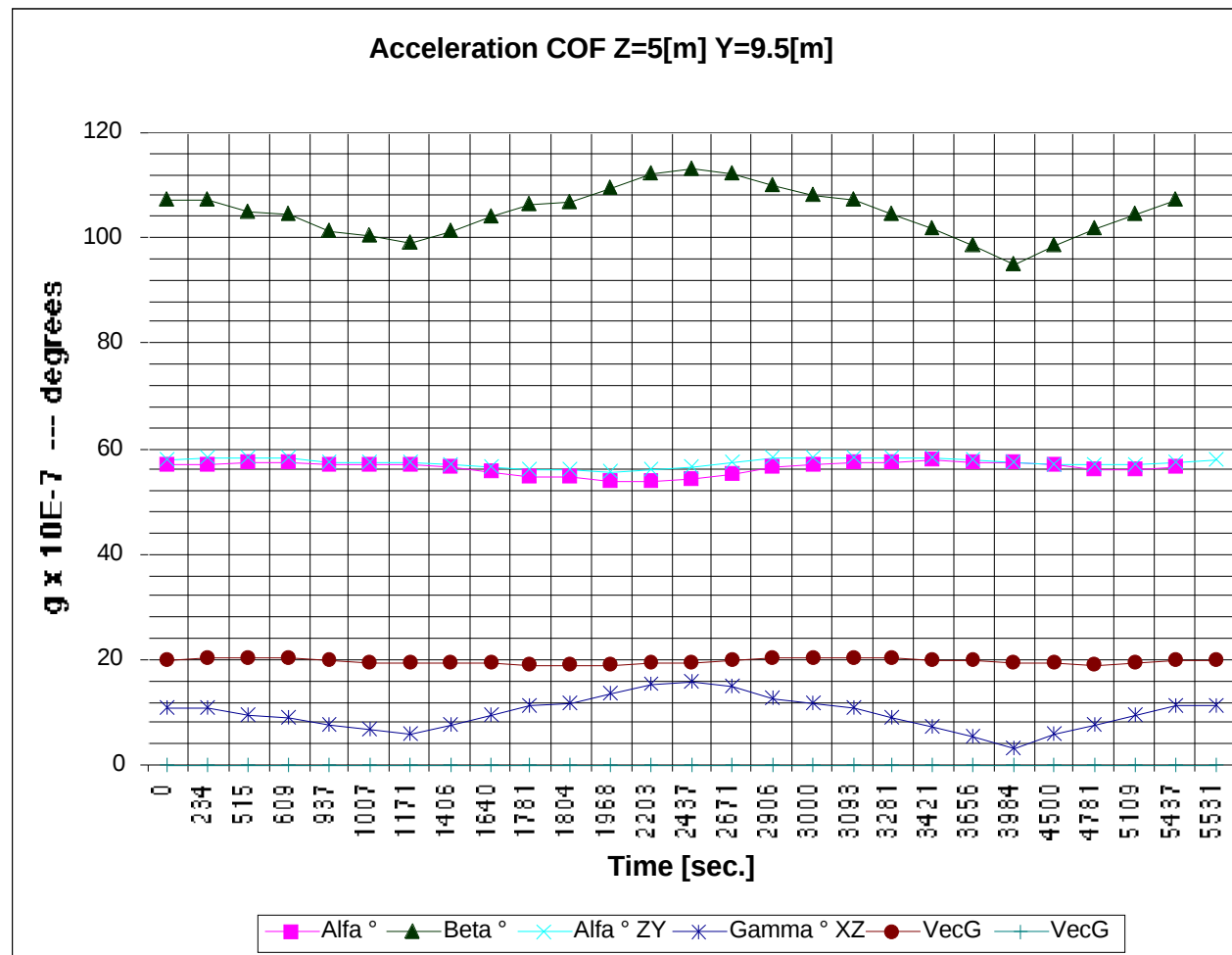
$H_{per} = 227.8 \text{ Nm}$
 $H_{apo} = 233.6 \text{ Nm}$



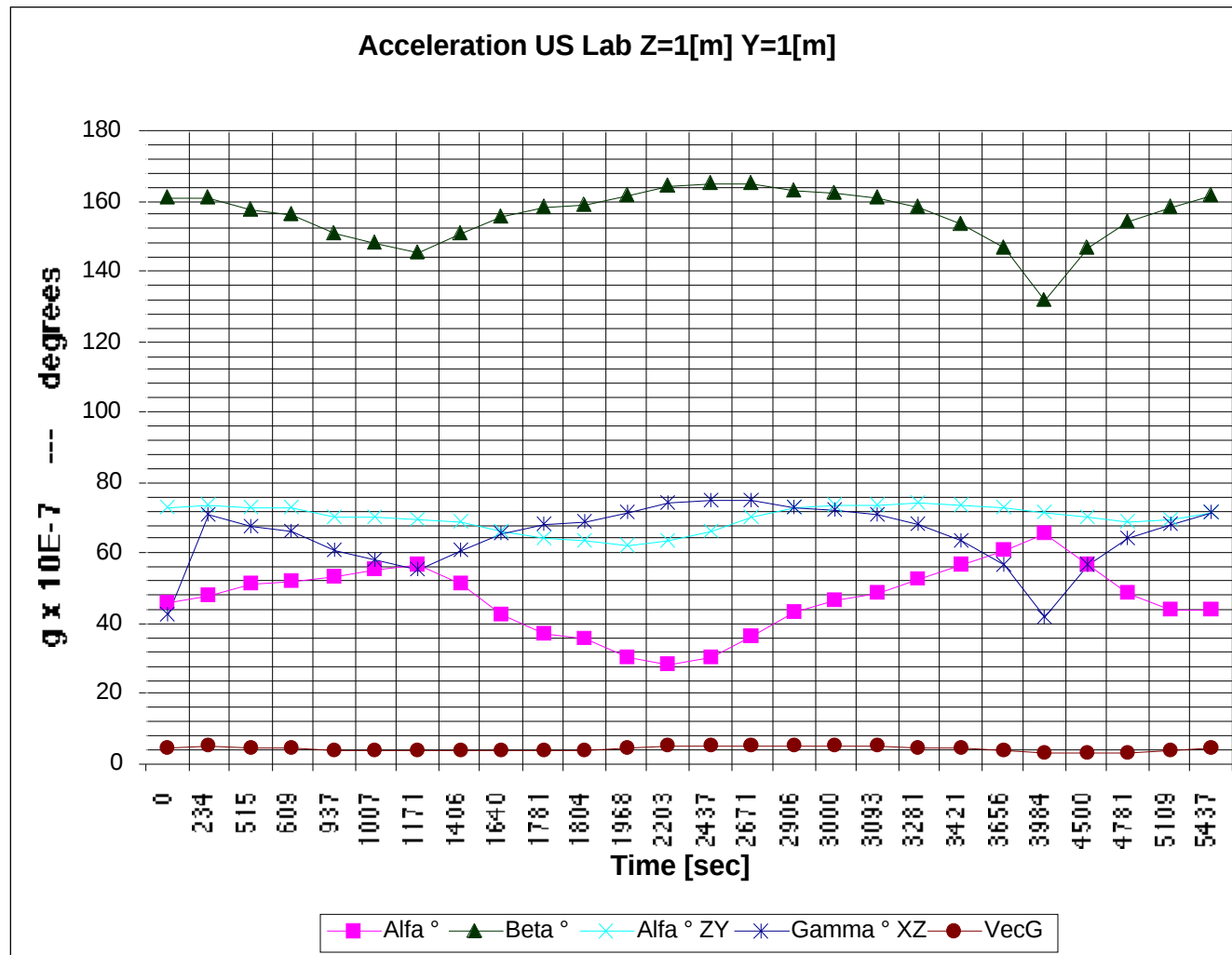
Source: Boeing

Variation range of residual-g directions and module during ISS Orbit (g-vectors are in scale)

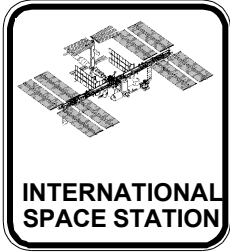




Time evolution of residual-g directions and module during ISS Orbit (inside COF)



Time evolution of residual-g directions and module during ISS Orbit (inside US Lab)



DAC5 Quasi-Steady Microgravity Assessment



DAC5 ASSEMBLY COMPLETE PERFORMANCE

Location (ISPRs)	Magnitude (μg)	Stability (μg)	Unit Vector Components			Angle (deg)	Location (Others)	Magnitude (μg)	Stability (μg)	Unit Vector Components			Angle (deg)
			X	Y	Z					X	Y	Z	
USL-C1	0.525	0.126	0.641	-0.109	0.760	17.370	USL-CG	0.793	0.137	0.466	-0.052	0.883	10.360
USL-C2	0.468	0.116	0.721	-0.133	0.680	20.710	APM-CG	1.624	0.157	0.288	-0.533	0.795	5.547
USL-C3	0.419	0.100	0.819	-0.165	0.549	24.820	APM-CLG1	1.010	0.151	0.350	-0.649	0.676	8.635
USL-C4	0.380	0.078	0.922	-0.204	0.330	28.370	APM-CLG2	1.120	0.154	0.313	-0.726	0.612	7.944
USL-C5	0.356	0.064	0.972	-0.237	-0.002	25.280	RS-FGB	1.119	0.139	-0.003	-0.060	-0.998	7.802
USL-S1	1.062	0.145	0.385	-0.227	0.895	7.927	RS-SM	2.179	0.129	-0.099	-0.038	-0.994	3.655
USL-S2	0.989	0.143	0.400	-0.248	0.883	8.421	JEM-CG	1.811	0.157	0.244	0.745	0.621	5.143
USL-S3	0.917	0.141	0.417	-0.272	0.867	8.973	JEF1-F1	2.954	0.165	0.223	0.627	0.746	3.325
USL-S4	0.846	0.138	0.437	-0.300	0.848	9.859	JEF2-A1	2.613	0.160	0.218	0.706	0.674	3.646
USL-P1	1.019	0.145	0.396	0.166	0.903	8.310	JEF3-F2	3.039	0.167	0.216	0.658	0.722	3.265
USL-P2	0.945	0.143	0.413	0.180	0.893	8.866	JEF4-A2	2.710	0.162	0.209	0.734	0.646	3.558
USL-P4	0.799	0.138	0.458	0.215	0.862	10.230	JEF5-F3	3.129	0.169	0.209	0.685	0.698	3.208
JPM1-A1	1.250	0.150	0.348	0.333	0.877	7.015	JEF6-A3	2.811	0.164	0.201	0.760	0.619	3.477
JPM2-F1	1.480	0.154	0.325	0.282	0.903	6.095	JEF7-F4	3.223	0.171	0.203	0.710	0.674	3.155
JPM3-A2	1.296	0.151	0.334	0.433	0.838	6.819	JEF8-A4	2.915	0.167	0.194	0.782	0.593	3.401
JPM4-F2	1.519	0.154	0.316	0.370	0.874	5.979	JEF9-O1	3.303	0.174	0.188	0.771	0.608	3.135
JPM5-A3	1.355	0.151	0.318	0.520	0.793	6.570	JEF10-O2	3.091	0.174	0.174	0.838	0.517	3.334
JPM6-F3	1.569	0.155	0.305	0.450	0.839	5.824	JEF11-U1	2.456	0.169	0.184	0.861	0.474	4.064
JPM7-A4	1.425	0.152	0.301	0.594	0.746	6.288	JEF12-U2	2.553	0.171	0.170	0.890	0.423	3.955
JPM8-A5	1.505	0.153	0.284	0.657	0.699	5.992	S3LO	3.299	0.223	0.038	-0.994	-0.104	3.918
JPM9-F5	1.700	0.156	0.280	0.584	0.762	5.441	S3LI	2.945	0.212	0.042	-0.991	-0.124	4.180
JPM10-F6	1.778	0.157	0.266	0.638	0.723	5.234	S3UO	3.958	0.209	-0.056	-0.846	-0.530	3.142
APM-FWD1	1.605	0.155	0.305	-0.386	0.871	5.573	S3UI	3.644	0.196	-0.062	-0.810	-0.584	3.222
APM-FWD2	1.681	0.157	0.291	-0.465	0.836	5.370	P3LO	3.260	0.191	0.022	0.973	-0.231	3.355
APM-FWD3	1.768	0.159	0.277	-0.532	0.800	5.167	P3UO	4.043	0.176	-0.068	0.780	-0.622	2.494
APM-FWD4	1.863	0.161	0.263	-0.590	0.763	4.968	10 ISPRs Have Quasi-Steady Magnitude $\leq 1.0 \mu\text{g}$						
APM-AFT1	1.397	0.152	0.318	-0.451	0.834	6.275							
APM-AFT2	1.482	0.154	0.300	-0.534	0.791	5.989							
APM-AFT3	1.578	0.157	0.282	-0.603	0.747	5.709							
APM-AFT4	1.682	0.160	0.264	-0.659	0.704	5.450							